

Optimal Control of Airborne Wind Energy Systems

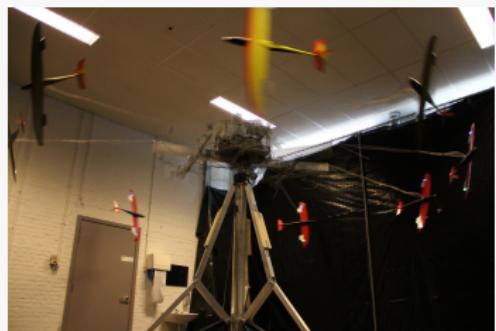
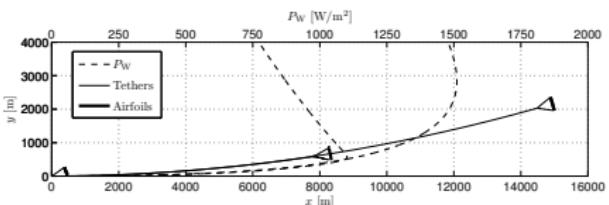
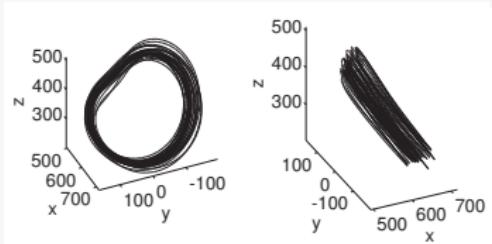
Greg and Mario

1 Control of Tethered Airfoils

1 Control of Tethered Airfoils

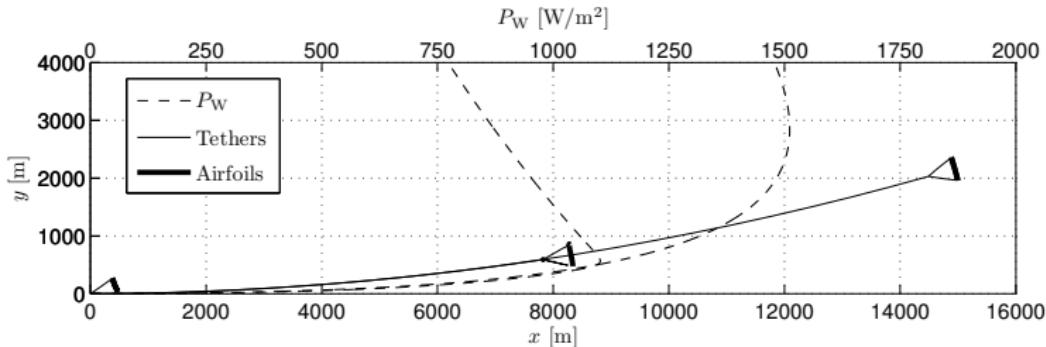
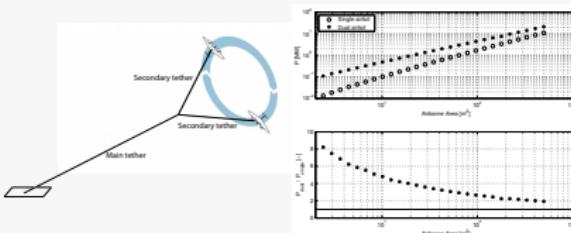
Control of Tethered Airfoils

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Dual Airfoils [Zanon et al. 2013]

- More advantageous than single airfoil
- More complex, nonlinear and unstable



Dual Airfoils

Pointmass model in spherical coordinates [Williams et al., 2008]

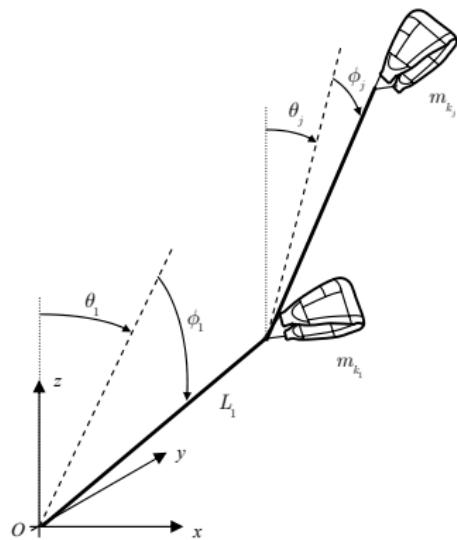


Fig. 2. Multiple kite model with kites on single line.

Spherical Coordinates

Dual airfoil model in spherical coordinates...

$$A = \begin{bmatrix} A_{11} & 0 & 0 & A_{14} & A_{15} & 0 & A_{17} & A_{18} & 0 \\ 0 & A_{22} & 0 & A_{24} & A_{25} & 0 & A_{27} & A_{28} & 0 \\ 0 & 0 & 1 & A_{34} & A_{35} & 0 & A_{37} & A_{38} & 0 \\ A_{41} & A_{42} & 0 & A_{44} & 0 & 0 & 0 & 0 & 0 \\ A_{51} & A_{52} & 0 & A_{54} & A_{55} & 0 & A_{57} & A_{58} & 0 \\ A_{61} & A_{62} & 0 & A_{64} & A_{65} & 0 & A_{67} & A_{68} & 0 \\ A_{71} & A_{72} & 0 & A_{74} & A_{75} & 0 & A_{77} & A_{78} & 0 \\ A_{81} & A_{82} & 0 & A_{84} & A_{85} & 0 & A_{87} & A_{88} & 0 \end{bmatrix} \quad (35)$$

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$$A_{11} = 1 / 3 L_0^{-2} \cos^2 \phi_0 \left(\rho_0 L_0 + 3 m_0 + 3 L_0 \rho_0 + 3 L_0 \rho_0 + 3 m_0 \right) \quad (36)$$

$$A_{\beta_1} = 1/2 L_i L_j \cos\phi_i \cos\phi_j (\sin\theta_i \sin\theta_j + \cos\theta_i \cos\theta_j) (L_i p_i + 2 m_i) \quad (37)$$

$$A_{11} = -1 / 2 L_0 L_1 \sin \phi_1 \cos \phi_0 (\sin \theta_1 \cos \theta_0 - \cos \theta_1 \sin \theta_0) (L_1 p_1 + 2 m_1) \quad (38)$$

$$A_{12} = 1 / 2 L_1 L_2 \cos\phi_1 \cos\phi_2 (\sin\theta_1 \sin\theta_2 + \cos\theta_1 \cos\theta_2) (L_2 p_2 + 2 m_2) \quad (39)$$

$$= -1 / 2 L_3 L_2 \sin \phi_2 \cos \phi_0 \left(\sin \theta_2 \cos \theta_0 - \sin \theta_0 \cos \theta_2 \right) \left(L_2 p_2 + 2 m_2 \right) \quad (40)$$

$$A_{22} = 1 / 3 L_0^{-1} \left(\rho_0 L_0 + 3 m_2 + 3 L_0 \rho_1 + 3 L_0 \rho_2 + 3 m_1 \right) \quad (41)$$

$$A_{21} = 1/2 L_0 L_1 \cos\phi_1 \sin\phi_0 (\sin\theta_1 \cos\theta_0 - \cos\theta_1 \sin\theta_0) (L_0 p_1 + 2 m_1) \quad (42)$$

$$+ \sin\phi_1 \sin\phi_0 \sin\theta_0 + \cos\phi_1 \cos\phi_0 + \sin\phi_1 \cos\theta_1 \sin\phi_0 \cos\theta_0 \left(L_1 \rho_1 + 2m_1 \right) \quad (45)$$

$$A_{2,2} = 1 / 2 L_0 L_2 \sin \phi_0 \cos \phi_2 (\sin \theta_0 \cos \theta_2 - \sin \theta_0 \cos \theta_2) (L_2 \beta_1 + 2 M_2) \quad (44)$$

$$v_{23} = v_1 + v_2 v_3 \left(\sin v_3 \cos v_2 \sin v_0 \cos v_1 + \sin v_3 \cos v_2 \cos v_0 \cos v_1 + \cos v_3 \cos v_2 \cos v_0 \right) \left(\sin v_2 + \sin v_3 \right) \quad (4.4)$$

$$t_1 + t_2 - 8 = t_1 + t_2 - 8 = t_1 + t_2 - 8 = t_1 + t_2 - 8 \quad (47)$$

$$t_1 = 1/(2L) - \epsilon - \epsilon \left(\frac{1}{2} + \theta - \theta_{\text{min}} - \theta_{\text{max}} \right) (L - \epsilon^2 \theta_{\text{min}}). \quad (48)$$

$$A_{\pm} = 1/(2L) \left(-\sin\phi \sin\theta \cosh\phi \sin\theta + \cos\phi \sin\phi - \sin\phi \cos\theta \cosh\phi \cos\theta \right) (L_{\pm} a_{\pm} + 2m_{\pm}) \quad (49)$$

$$A_1 = 1/3 U^2 \cos^2 \phi \left(k_1 a + 3 m \right) \quad (50)$$

$$A_{\perp} = 1/3 L_1^2 \left(L_1 p_z + 3m_e \right) \quad (51)$$

$$A_{\pm 1} = -L_0 \cos\phi_0 \sin\phi_0 (\cos\theta_1 \sin\theta_0 - \cos\theta_0 \sin\theta_1) (L_0 p_1 + m_1) \quad (52)$$

$$\cos\phi_1 \sin\theta_1 \sin\phi_0 \sin\theta_0 - \cos\phi_1 \cos\theta_1 \sin\phi_0 \cos\theta_0 \right) (L_1 p_1 + m_1) \quad (53)$$

$$A_{i,j} = 1/3 L_i^2 \cos^2 \phi_j (L_j \rho_i + 3 m_j) \quad (54)$$

$$A_{i,i} = 1 / 3 L_2 \cdot \left(L_3 \rho_2 + 3 m_3 \right) \quad (55)$$

$$\tilde{\phi}_2 \cos \tilde{\phi}_0 \left(\sin \tilde{\theta}_2 \cos \tilde{\theta}_0 - \cos \tilde{\theta}_2 \sin \tilde{\theta}_0 \right) \left[L_2 \rho_2 + m_2 \right] \quad (36)$$

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$$A_{\perp} = L \left(\sin \phi \cos \phi, -\cos \phi \sin \theta, \sin \phi \sin \theta, -\cos \phi \cos \theta \sin \phi, \cos \theta \right) / (L\rho + m) \quad (57)$$

The right-hand side of the equations of motion are given by

$$\begin{aligned}
& + 1/2 \cdot L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \partial_{\alpha_1}^2 L_2 \\
& + 2 \cdot L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \sin(\alpha_1) \partial_{\alpha_1} \partial_{\alpha_2} L_2 \\
& - 1/2 \cdot L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \cos(\alpha_1) \partial_{\alpha_1}^2 L_2 \\
& - 2 \cdot L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \sin(\alpha_1) \partial_{\alpha_1} \partial_{\alpha_2} L_2 \\
& - 2 \cdot L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \cos(\alpha_1) \partial_{\alpha_1} \partial_{\alpha_2} L_2 \\
& - L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \sin(\alpha_1) \partial_{\alpha_1} L_2 \\
& - L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \cos(\alpha_1) \partial_{\alpha_1} L_2 \\
& - L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \sin(\alpha_1) \partial_{\alpha_1} \partial_{\alpha_2} L_2 \\
& - 2 \cdot L_1^2 \cdot \cos(\alpha_1) \sin(\alpha_1) \cdot [L_1^2 \cdot L_2^2 \cdot \cos(\alpha_2) \sin(\alpha_2) - 2 \cdot L_1^2 \cdot L_2 \cdot \cos(\alpha_2) \sin(\alpha_2)] \cdot \cos(\alpha_1) \partial_{\alpha_1} \partial_{\alpha_2} L_2
\end{aligned} \tag{58}$$

Control of Tethered Airfoils

Spherical Coordinates

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... pages 5 & 6 ...

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Spherical Coordinates

...page 7.

$$\begin{aligned}
 \dot{\theta}_1 &= -2m_1 \dot{L}_1 \dot{\phi}_1 \dot{\theta}_2 - 2m_1 L_1 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 \cos(\theta_1) \dot{\theta}_2 \\
 &\quad - 2m_1 \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \cos(\theta_1) \dot{\theta}_2^2 \sin(\phi_1) \dot{\theta}_2 \\
 &\quad - m_1 \dot{L}_1 \cos(\phi_1) \dot{L}_1 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_2 \cos(\phi_1) \dot{\theta}_2 \\
 &\quad - m_1 \dot{L}_1 \cos(\phi_1) \dot{L}_1 \sin(\phi_1) + 1/2 \rho_1 g_1 \sin(\phi_1) \cos(\theta_1)^2 + 2m_1 \dot{L}_1 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_2 \\
 &\quad + m_1 \dot{L}_1 \cos(\phi_1) \dot{L}_1 \sin(\phi_1) \dot{\theta}_2^2 - 1/2 \rho_1 g_1 \cos(\phi_1)^2 - m_1 \dot{L}_1 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_2 \\
 &\quad - m_1 \dot{L}_1 \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\phi}_1 \cos(\phi_1)^2 - m_1 \dot{L}_1 \cos(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_2 \\
 &\quad + 2m_1 \dot{L}_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\phi}_1 \sin(\phi_1) \dot{\theta}_2^2 + 1/2 \rho_1 g_1 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_2 \\
 &\quad - m_1 \dot{L}_1 \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\phi_1)^2 + m_1 \dot{L}_1 \cos(\phi_1) \cos(\phi_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 \sin(\phi_1) \dot{\theta}_2 \\
 &\quad + 1/2 m_1 \dot{L}_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_2^2 - 1/2 \rho_1 g_1 \cos(\phi_1)^2 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_2^2 \\
 &\quad - 1/2 \rho_1 g_1 \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\phi}_1 \sin(\phi_1) \dot{\theta}_2^2 - 1/2 \rho_1 g_1 \cos(\phi_1)^2 \sin(\phi_1) \dot{\theta}_2^2 \\
 &\quad - 1/2 \rho_1 g_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\phi}_1 \cos(\phi_1)^2 + 1/2 \rho_1 g_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_2^2 \\
 &\quad - m_1 \dot{L}_1 \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\phi_1)^2 + m_1 \dot{L}_1 \cos(\phi_1) \cos(\phi_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 \sin(\phi_1) \dot{\theta}_2 \\
 &\quad + 1/2 \rho_1 g_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\phi_1)^2 - 1/2 \rho_1 g_1 \cos(\phi_1)^2 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_2^2 \\
 &\quad - 1/2 \rho_1 g_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\phi_1)^2 - m_1 \dot{L}_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\phi}_1 \sin(\phi_1) \dot{\theta}_2 \\
 &\quad - m_1 \dot{L}_1 \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\phi_1)^2 - m_1 \dot{L}_1 \cos(\phi_1) \cos(\phi_1) \dot{L}_1 \sin(\phi_1) \cos(\phi_1) \dot{\theta}_2 \\
 &\quad + 2m_1 \dot{L}_1 \sin(\phi_1) \cos(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\phi_1)^2 + Q_1 \\
 \dot{\theta}_2 &= -\rho_1 \dot{L}_1 \dot{\theta}_1 + 1/2 \rho_1 \dot{L}_1 \dot{\phi}_1^2 + 1/2 \rho_1 \dot{L}_1 \dot{\theta}_1^2 - m_1 \cos(\phi_1) \cos(\theta_1) \\
 &\quad + m_1 \rho_1 \dot{L}_1 \sin(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 + m_1 \cos(\phi_1) \cos(\theta_1) \cos(\phi_1) \cos(\theta_1) \dot{\theta}_1^2 \\
 &\quad + m_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\phi}_1^2 - 2m_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 \dot{\phi}_1 \\
 &\quad + 2m_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 + m_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 \dot{\phi}_1 \\
 &\quad - 2m_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_1 - m_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 \cos(\phi_1) \dot{\theta}_1 \\
 &\quad + 2m_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 - m_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 \sin(\phi_1) \dot{\theta}_1 \\
 &\quad - \rho_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_1 + \dot{\rho}_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_1 \\
 &\quad + \rho_1 \sin(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 - 2\rho_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 \\
 &\quad - 2m_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 - m_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_1 \\
 &\quad + 2m_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 - m_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_1 \\
 &\quad + \rho_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 + \dot{\rho}_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \cos(\phi_1) \cos(\theta_1) \dot{\theta}_1 \\
 &\quad + 2\rho_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 + \rho_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 \\
 &\quad - 2\rho_1 \cos(\phi_1) \cos(\theta_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 \dot{\theta}_1 + 1/2 \rho_1 \dot{L}_1 \sin(\phi_1) \dot{\theta}_1^2 \\
 &\quad + 1/2 \rho_1 \dot{L}_1 \sin(\phi_1) \dot{\theta}_1^2 + 2\rho_1 \cos(\phi_1) \sin(\phi_1) \dot{L}_1 \sin(\phi_1) \dot{\phi}_1 \dot{\theta}_1 - m_1 \sin(\phi_1) \dot{L}_1 \sin(\phi_1) \\
 &\quad - m_1 \dot{L}_1 + Q_2 - \rho_1 \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_1 - \rho_1 \dot{L}_1 \cos(\phi_1) \sin(\phi_1) \dot{\theta}_1 - \rho_1 \sin(\phi_1) \dot{L}_1 \sin(\phi_1) \dot{\theta}_1 \\
 &\quad - 2m_1 \sin(\phi_1) \dot{L}_1 \cos(\phi_1) \dot{\theta}_1 + m_1 \sin(\phi_1) \dot{L}_1 \sin(\phi_1) \dot{\theta}_1 - 1/2 \rho_1 \dot{L}_1 + 1/2 \rho_1 \dot{L}_1
 \end{aligned} \tag{66}$$

The above equations of motion are implemented in MATLAB and solved as a function of time by expressing them in state-space form.

C. External Forces

1. Aerodynamic Kite Forces
 The aerodynamic forces acting on the kite-torsion system that are not modeled thus far in the equations of motion are the lift and drag forces from the kite, together with the drag forces on the tether. The kite is assumed to be controlled by manipulating its angle of attack and roll angle. Thus, in this study, its attitude dynamics are ignored. The lift and drag forces due to the kite are derived using a velocity coordinate system, as shown in 4.

Cartesian Coordinates

$$\begin{bmatrix} M & C \\ C^T & 0 \end{bmatrix} \begin{bmatrix} \ddot{\vec{r}} \\ \ddot{\vec{\omega}} \\ \nu \end{bmatrix} = F, \quad \ddot{\vec{r}} = \begin{bmatrix} \ddot{\vec{r}}_0 \\ \ddot{\vec{r}}_1 \\ \ddot{\vec{r}}_2 \end{bmatrix}, \quad \ddot{\vec{\omega}} = \begin{bmatrix} \dot{\vec{\omega}}_1 \\ \dot{\vec{\omega}}_2 \end{bmatrix}, \quad \ddot{\vec{\nu}} = \begin{bmatrix} \nu_0 \\ \nu_1 \\ \nu_2 \end{bmatrix}$$

$$M = \begin{bmatrix} \sum_{k=0}^3 \xi_k & \frac{1}{2} \xi_1 & \frac{1}{2} \xi_2 & 0 & 0 \\ \frac{1}{2} \xi_1 & \xi_1 + m_1 I_3 & 0 & 0 & 0 \\ \frac{1}{2} \xi_2 & 0 & \xi_2 + m_2 I_3 & 0 & 0 \\ 0 & 0 & 0 & J_1 & 0 \\ 0 & 0 & 0 & 0 & J_2 \end{bmatrix}, \quad C = \begin{bmatrix} \vec{r}_0 & \nabla_{\vec{r}_0} c_1 & \nabla_{\vec{r}_0} c_2 \\ 0 & \nabla_{\vec{r}_1} c_1 & 0 \\ 0 & 0 & \nabla_{\vec{r}_2} c_2 \\ 0 & 2P_{R_1} (\nabla_{R_1} c_1) & 0 \\ 0 & 0 & 2P_{R_2} (\nabla_{R_2} c_2) \end{bmatrix},$$

$$F = \begin{bmatrix} \vec{F}_0 - \frac{1}{2} g \mu_0 l_0 \vec{l}_3 - \sum_{k=1}^2 \frac{1}{2} g \mu_k l_k \vec{l}_3 \\ \vec{F}_1 - \frac{1}{2} g \mu_1 l_1 \vec{l}_3 - g m_1 \vec{l}_3 \\ \vec{F}_2 - \frac{1}{2} g \mu_2 l_2 \vec{l}_3 - g m_2 \vec{l}_3 \\ \vec{M}_1 - \omega_1 \times J_1 \omega_1 \\ \vec{M}_2 - \omega_2 \times J_2 \omega_2 \\ \nabla_{\vec{r}_0} \dot{c}_0^T \vec{r}_0 \\ -\nabla_{\vec{r}_0} \dot{c}_1^T \vec{r}_0 - \nabla_{\vec{r}_1} \dot{c}_1^T \vec{r}_1 - 2P_{R_1} (\nabla_{R_1} \dot{c}_1)^T \omega_1 \\ -\nabla_{\vec{r}_0} \dot{c}_2^T \vec{r}_0 - \nabla_{\vec{r}_2} \dot{c}_2^T \vec{r}_2 - 2P_{R_2} (\nabla_{R_2} \dot{c}_2)^T \omega_2 \end{bmatrix}$$

$$\dot{c}_0 = \vec{r}_0^T \dot{\vec{r}}_0, \quad \dot{c}_k = (\vec{r}_k + R_k \vec{r}_T - \vec{r}_0)^T (\dot{\vec{r}}_k + R_k \omega_k \times \vec{r}_T - \dot{\vec{r}}_0)$$

$$\nabla_{\vec{r}_0} c_0 = \vec{r}_0, \quad -\nabla_{\vec{r}_0} \dot{c}_k = \nabla_{\vec{r}_k} \dot{c}_k = (\dot{\vec{r}}_k + R_k \omega_k \times \vec{r}_T - \dot{\vec{r}}_0),$$

$$2P_{R_k} (\nabla_{R_k} c_k) = \vec{r}_T \times R_k^T (\vec{r}_k - \vec{r}_0), \quad -\nabla_{\vec{r}_0} c_k = \nabla_{\vec{r}_k} c_k = \vec{r}_k + R \vec{r}_T - \vec{r}_0,$$

$$2P_{R_k} (\nabla_{R_k} \dot{c}_k) = R^T (\vec{r} - \vec{r}_0) \times (\omega \times \vec{r}_T) + R^T (\dot{\vec{r}} - \dot{\vec{r}}_0) \times \vec{r}_T,$$

Dual Airfoils

Airfoil model: index-1 DAE

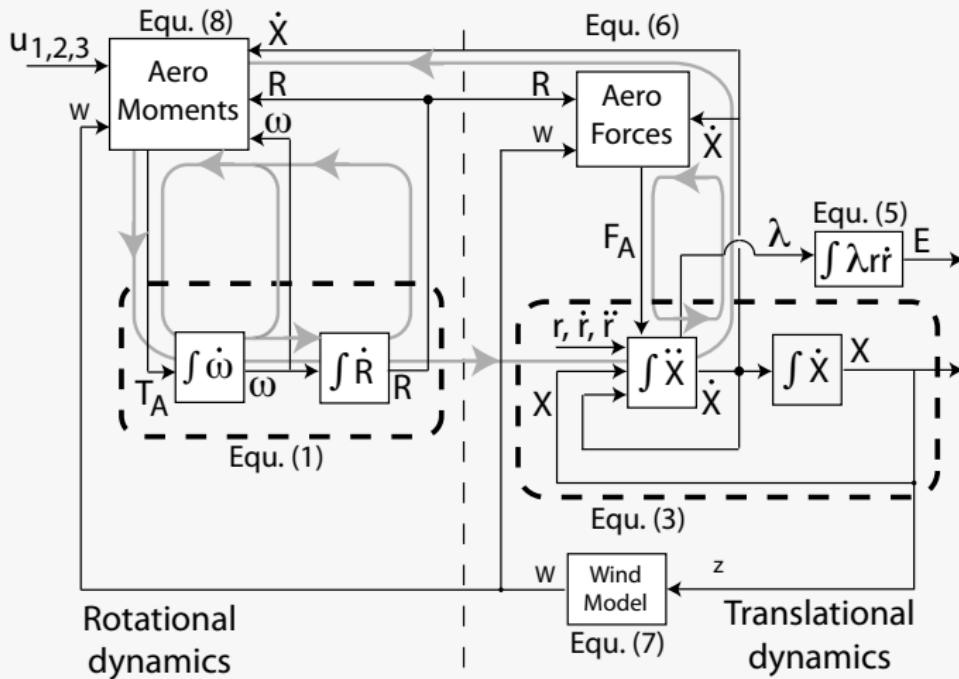
- 50 differential states
- 3 algebraic states
- 8 controls

Wind turbulence model:

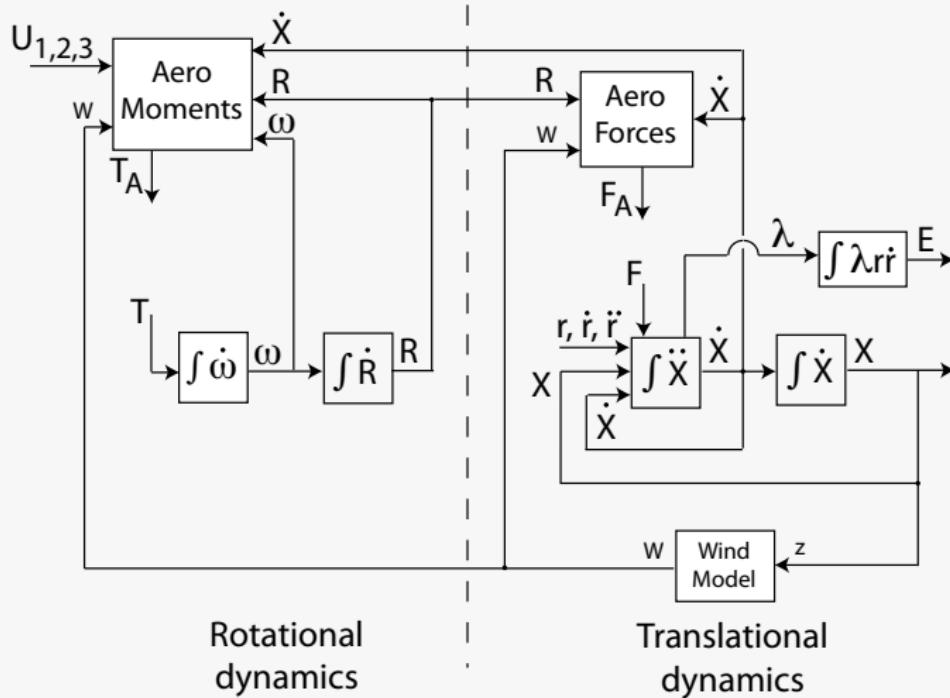
$$\dot{w}_{\diamond \star} = -\frac{w_{\diamond \star}}{\tau} + u_{\diamond \star}, \quad \diamond \in \{x, y, z\}, \star \in \{1, 2\}$$

- 6 differential states
- 6 controls

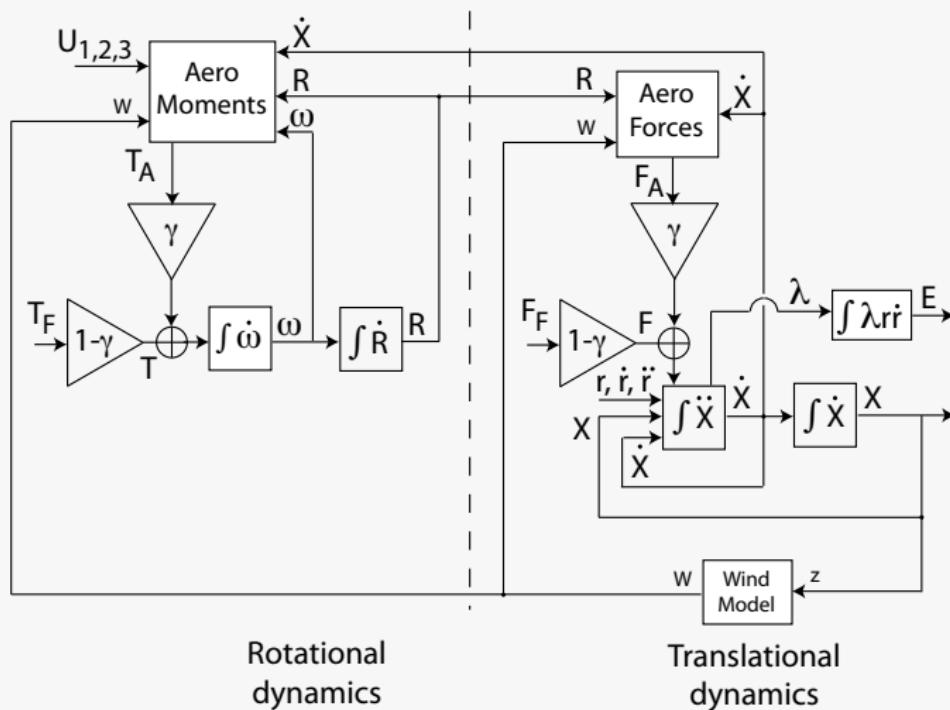
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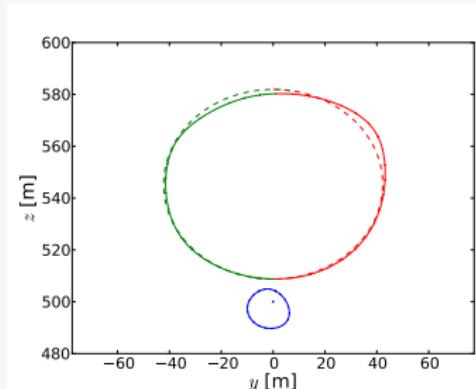
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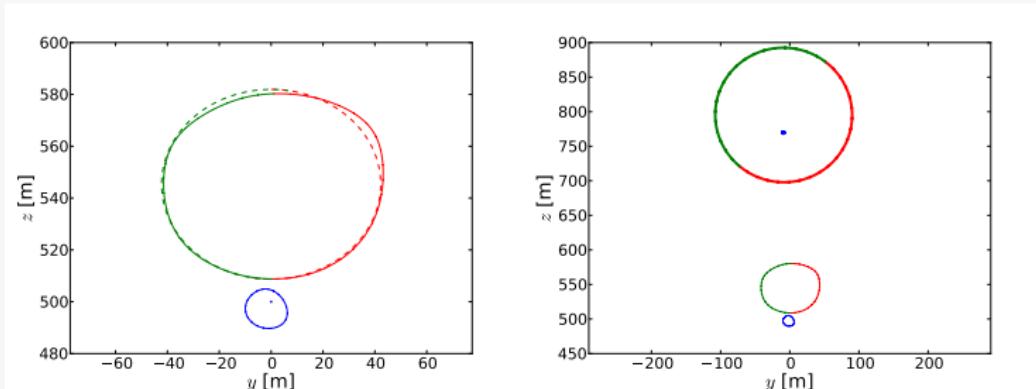
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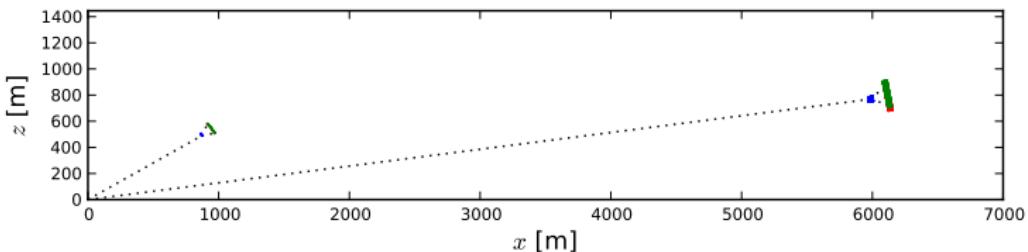
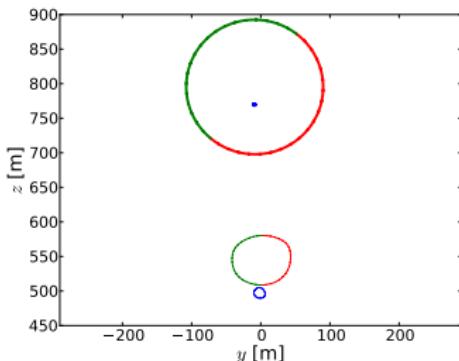
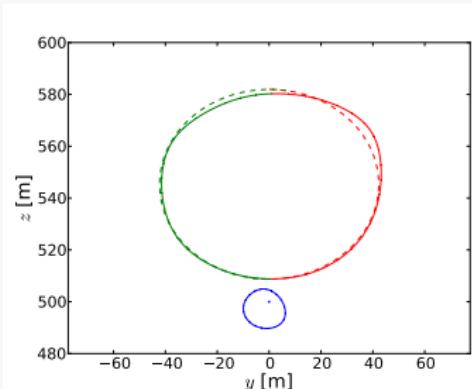
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